

T = engine thrust.

L = total wing lift plus fuselage lift.

D = total airplane drag.

in = moment of L and D with reference to wing a.c. (aerodynamic center)

W = weight of airplane.

, = inertia force normal to flight path.

n = inertia force parallel to flight path.

= rotation inertia moment.

g = tail load normal to flight path.

For a horizontal constant velocity flight condition, the inertia forces  $I_1$ ,  $I_n$ , and  $I_m$ could be zero. For an accelerated flight conition involving translation but not angular scoeleration about its own c.g. axis, the inertia moment I\_ would be zero, but I, and  ${
m I_{m}}$ would have values.

## Equations of Equilibrium For Steady Flight.

From Fig. A4.10 we can write:-

EF\_ = o, D + W sin Q-T cos B = o

L - W cos O+T sin B - E = o

- Ma - La - Db + Tc cos 8 + Es = o  $\mathbf{z}_{\mathbf{x}_{i,j}} = \mathbf{o}_{i,j}$ 

Equations of Equilibrium in Accelerated Flight.

 $ZF_{\omega} = o_{\star}$ D + W sin @ - T cos B - In = 0

EF\_ = 0, L = W cos 9 + T sin B = I, = E = 0

ZM = 0, - Ma - La - Db + Tc cos β + Ee + I = 0

Forces - Plus is up and toward tail. Signs used: | Moment - Clockwise is positive.

Distances from c.g. to force -

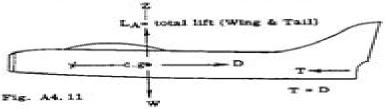
Plus is up and toward tail.

#### A4.7 Load Factors.

The term <u>load factor</u> normally given the symbol (n) can be defined as the numerical mul-

. The bar through letter Z has no significance. Same meaning without bar.

tiplying factor by which the forces on the airplane in steady flight are multiplied to obtain a static system of forces equivalent to the dynamic force system acting during the acceleration of the airplane. Fig. A4.11 illustrates



forces in steady horizontal flight. L represents the total airplane lift (wing plus tail). Therefore L = W. Now assume the airplane is accelerated upward slong the Z axis. Fig. A4.12 shows the additional inertia force Wa\_/g acting downward, or opposite to the direction of acceleration. The total sirplane lift L for the



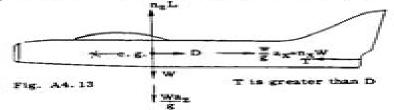
unaccelerated condition in Fig. A4.11 must be multiplied by a load factor  $n_{\alpha}$  to produce stati. equilibrium in the 2 direction.

Thus, 
$$n_a L - W - \frac{W}{K} a_a = 0$$

Since L = W

Hence 
$$n_{\alpha} = 1 + \frac{n_{\beta}}{\sigma}$$

An airplane can of course be accelerated along the K axis as well as the Z axis. Thus in Fig. A4.13 the magnitude of the engine thrust T is greater than the airplane Drag D, which



causes the airplane to accelerate forward. It convenient to express the inertia force in the direction in terms of the load factor n and the

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